

IN THE CLAIMS

Please amend the claims as follows:

1-16 (Cancelled)

17. (Currently Amended) An aircraft fuselage, having a fuselage structure, comprising:

an exterior skin which is designed to be part of the structural elements that are components of the mechanical strength bracing of the fuselage and help absorb its forces and is made of materials designed to be resistant to shear, and incorporated as a bearing element into the mechanical strength bracing to absorb and transfer the forces and torques acting thereupon, wherein the exterior skin comprises a composite material and a metallic material, such that the exterior skin is a hybrid material capable of being molded and joined by further processing, wherein the composite material comprises carbon fibers or combinations of carbon fibers with glass fibers or ceramic fibers, and the carbon fibers are coated with a nitride or a carbide bond and are embedded in a metal or a ceramic such that the composite material is heatproof and wherein the metallic material is selected from the group of metallic materials consisting of an aluminum, a titanium, an aluminum alloy, a titanium alloy, and combinations thereof, wherein the composite material and the metallic material are coated by a resin layer or are embedded in a resin, wherein the exterior skin has a sandwich design, in which the composite material and the metal material are adhesively bonded in layers, wherein the sandwich design is adapted for yielding a burn-through resistant behavior of the exterior skin relative to extended exposure to flames from a fire.

18. (Cancelled)

19. (Previously presented) The aircraft fuselage of claim 17, wherein the carbon fibers coated with a nitride or carbide bond are embedded in a ceramic.

20. (Cancelled)

21. (Previously presented) The aircraft fuselage of claim 17, wherein the composite material includes plastics reinforced with glass or plastic fibers.
22. (Previously presented) The aircraft fuselage of claim 17, wherein the metallic material is made of titanium or of a titanium alloy.
23. (Previously presented) The aircraft fuselage of claim 17, wherein the composite material is comprised of a carbon fiber composite (CFK) material.
24. (Cancelled)
25. (Cancelled)
26. (Previously presented) The aircraft fuselage of claim 17, wherein the sandwich design includes a GLARE material, having a high burn-through behavior.
27. (Previously presented) The aircraft fuselage of claim 17, wherein an outer surface of the exterior skin exposed to weathering is protected by joining a plate-like planking to the outer surface, the planking being comprised of a non-metallic material, a fireproof metallic material, or a combination material, the combination material being comprised of a non-metallic material and a metal material, such that the planking is capable of being molded and joined by further processing.
28. (Previously presented) The aircraft fuselage of claim 27, wherein the planking is formed such that the planking is protective against burn through and adjusted to an outer contour of the exterior skin.
29. (Previously presented) The aircraft fuselage of claim 28, wherein the planking is realized using a GLARE material.

30. (Previously presented) The aircraft fuselage of claim 29, wherein the metallic material of the exterior skin comprises aluminum or an aluminum alloy joined to the planking.

31. (Currently Amended) The aircraft fuselage of claim 17, wherein the composite material is further comprised of a glass fiber composite (GSK) (GFK) material.

32. (Currently Amended) The aircraft fuselage of claim 23, wherein the composite material is further comprised of a glass fiber composite (GSK) (GFK) material.